

<b>7. MATERIALS USED IN PRIMARY STRUCTURE</b>			
<b>COMPONENT</b>		<b>DIMENSIONS</b>	<b>MATERIAL SPECIFICATION</b>
Mainplanes	Spars	63.5 mm x 2.5 mm	Aluminium 6061 - T8
	Ribs	6 mm & 4mm	Birch ply wood
	Covering	2.2 oz	Fabric
	Struts	25, 4 od x 0,9 mm	SAE 1010
Fuselage		10- 25,4 mm	SAE 1010
Tailplane	Spars	15,88 x 1.6 mm	SAE1010
	Covering	2.2 oz	Fabric
Fin	Spars	15,88 x 1.6 mm	SAE1010
	Ribs	10 x 0.9 mm	SAE1010
	Covering	2.2 oz	Fabric
Engine Mount		19 x 0.9 mm	SAE 1010
Control Surfaces			
Ailerons Flapperons	Spars	19 x 1.6 mm	Aluminium 6062 - T6
	Ribs	1,5 mm	Composite
	Covering	1 mm	Composite
Elevator	Spars	15,88 mm x 1,6 mm	SAE1010
	Ribs	10 x 0,9 mm	SAE1010
	Covering	2.2 oz	Fabric
Under-Carriage	Struts	25,4 x 0,9 mm	SAE1010
	Wheels	15 x 600 x 6	Pneumatic
	Axle	16mm	EN19
Fittings, rotor head fitting	Wingroot		
	Control attachment		
	Undercarriage attachment		
	Other		
Flying Controls	Cables	3mm 7 x 19 strand	Stainless Steel
	Push-pull rods	12,7 x 0,9 mm	SAE 1010
Miscellaneous Adhesives and glues dope		AERO-MUTI Covering system	
<b>7. OTHER (specify)</b>			
7.1 Indicate with an asterisk (*) which of the materials listed above is not covered by a recognized aircraft specification.			
7.2 Is a report of its physical characteristics attached to this application in respect of each material marked above with an asterisk?			
7.3 Copies of the following reports of the results of structural tests and any other tests (excluding flying) done to establish the integrity of the aircraft are attached to this application:			